

































Quadcopters

 Translational dynamics (Newton's law – includes mass/acceleration/ forces) (Gravity – Total thrust (rotated to the world frame)

$$m\dot{v} = \begin{bmatrix} 0\\0\\mg \end{bmatrix} - R_B^0 \begin{bmatrix} 0\\0\\T \end{bmatrix}, \quad T = \sum_{i=1..4} T_i$$

- Rotations are generated by pairwise differences in rotor thrusts (*d* distance from the center)
- Rolling and pitching torques around *x* and *y*
- Torque in *z* yaw torque

$$\begin{aligned} & Q_i = k\omega_i^2 & \text{Torque applied by the motor as opposed to} \\ & \sigma_{\text{aerodynamic drag}} & \tau_x = dT_4 - dT_2 \\ & \tau_x = db(\omega_4^2 - \omega_2^2) \\ & \tau_y = db(\omega_1^2 - \omega_3^2) \end{aligned}$$

Quadcopter dynamics	
• Rotational Dynamics, rot. acceleration in the airframe, Euler's eq. of motion $J\dot{\omega} = -\vec{\omega} \times J\vec{\omega} + \Gamma, \Gamma = [\tau_x, \tau_y, \tau_z]^T$	
 Where <i>J</i> is 3x3 inertia matrix Rotational Inertia of a body in 3D is represented by a 3x3 symmetric matrix J 	
 Diagonal elements are moments of inertia and off-diagonal are products of inertia Inertia matrix is a constant and depends on the mass and the shape of the body 	
$J = \begin{bmatrix} J_{xx} & J_{xy} & J_{xz} \\ J_{xy} & J_{yy} & J_{yz} \\ J_{xz} & J_{yz} & J_{zz} \end{bmatrix}$	















































